

Title of the invention

Making turbomachine turbines having blade inserts with resonant frequencies that are adjusted to be different, and a method of adjusting the resonant frequency of a 5 turbine blade insert

Background of the invention

The invention applies to turbomachines, and in particular to aviation turbines and to industrial gas 10 turbines. More particularly, the invention relates to the turbines of such turbomachines, and more precisely still, it relates to adjusting the resonant frequency of the blade inserts of such turbines.

Usually, the blade inserts of a turbine comprise a 15 single blade connected both to an outer platform and to an inner platform, which inner platform is in turn extended by a root for fixing the insert to the turbine rotor.

It is important to ensure that undesirable 20 vibrations do not appear in the turbines of turbomachines, and in particular vibrations caused by resonance phenomena, at different operating speeds of the turbomachine.

Unfortunately, given constraints in terms of 25 complying with the required aerodynamic profile for blades and complying with weight and size constraints, the options available for adjusting the resonant frequency of a blade insert by acting on its shape are limited.

30 Thus, French patent document FR 1 578 562 proposes giving the single-blade inserts of a wheel different characteristics such as roots of different lengths in the radial direction or different dimensions at the levels where the inserts are attached to the rotor. 35 Nevertheless, such solutions are penalizing in terms of insert weight and in terms of complexity in building the rotor or in attaching inserts to the rotor.

Object and brief summary of the invention

An object of the invention is to provide a method of making a turbomachine turbine that makes it possible to
5 avoid undesirable vibrations appearing.

Another object of the invention is to propose a method that provides great latitude in adjusting the resonant frequency of a turbine blade insert without penalizing it from the point of view of its aerodynamic
10 characteristics or its weight and without introducing complication in mounting blade inserts or in building the rotor.

Another object of the invention is to propose a turbine for a turbomachine in which the risk of
15 undesirable vibration appearing is greatly decreased, but without degrading turbine performance.

In a first aspect, the invention provides a method of making a turbomachine turbine, the turbine comprising at least one moving wheel provided with blade inserts, in
20 which method, each of the inserts used for the or each moving wheel comprises at least two blades interconnected by common inner and outer platforms and sharing a common root, at least some of the inserts having a hollow root in which a recess is formed, insert roots belonging to
25 the same wheel and/or insert roots belonging to two different wheels being voluntarily given different configurations for the recesses in their roots so as to adjust the resonant frequencies of said blade inserts to values that are significantly different, thereby ensuring
30 that the blade inserts of a single wheel and/or between two different wheels are out of tune.

In comparison with a wheel built up in conventional manner from single-blade inserts, the use of a root that is shared by a plurality of blades provides greater
35 options for acting on the configuration of a recess in the root for the purpose of adjusting the resonant frequency of a blade insert.

It is thus easier to ensure that the resonant frequencies of the inserts in a single moving wheel are out of tune or that the resonant frequencies of blade inserts belonging to different moving wheels are out of 5 tune, thereby reducing the risk of vibration appearing.

In another of its aspects, the invention provides a method of adjusting the resonant frequency of a moving blade insert for a turbomachine turbine, in which method the insert is made to have at least two blades 10 interconnected by common inner and outer platforms and sharing a common root, and the resonant frequency of the insert is adjusted by forming a recess in the insert root.

In particular, it is possible to make the blade 15 insert with a root in the form of a bulb and to adjust its resonant frequency by acting on the thickness of material on either side of the recess level with a root portion that is neck-shaped.

Advantageously, for a turbine blade insert for use 20 in a turbomachine having a high-pressure compressor (HP) and a low-pressure compressor (LP), the resonant frequency of each insert is adjusted to a value that is at least 14% greater than the excitation mode corresponding to $N_2 - N_1$ where N_2 and N_1 are the speeds of 25 rotation of the HP and LP compressors.

In yet another aspect of the invention, the invention provides a staged turbine for a turbomachine, which turbine comprises at least one moving wheel fitted with blade inserts each comprising at least two blades 30 interconnected by common outer and inner platforms and sharing a common root, in which turbine at least some of the inserts of the same moving wheel and/or of different moving wheels have hollow roots in which respective recesses are formed presenting configurations that differ 35 so that the inserts have resonant frequencies that are significantly different.

The recess formed in a blade insert root may open out to at least one side of the root, and it may be a blind recess.

Some of the blade inserts may have a root that is 5 solid without any insert.

Brief description of the drawings

The invention will be better understood on reading 10 the following description given by way of non-limiting indication with reference to the accompanying drawings, in which:

- Figure 1 is a fragmentary diagrammatic section view of a portion of a turbomachine;
- Figure 2 is a view on a larger scale showing part 15 of Figure 1;
- Figure 3 is a diagrammatic perspective view of a turbine blade insert applying the invention;
- Figure 4 is a diagrammatic view on a larger scale of the inside end portion of the Figure 3 insert;
- 20 - Figure 5 is a diagrammatic view on a larger scale showing the side-by-side disposition of the inside end portions of two inserts as shown in Figure 3 disposed side by side in a moving wheel of a turbine;
- Figure 6 is a graph plotting curves representing 25 the resonant frequency of an insert as shown in Figure 3 and of a single-blade insert, as a function of speed of rotation; and
- Figure 7 is a diagrammatic view on a larger scale 30 showing the inside end portion of a moving blade insert in another embodiment of the invention.

Detailed description of embodiments

Figure 1 is a highly diagrammatic meridian half-section view of a turbomachine turbine, e.g. in an 35 aircraft turbojet.

In well-known manner, the turbine 10 comprises a high-pressure turbine portion (HP turbine) 12 coupled to

a shaft (not shown) for rotating a high-pressure compressor (HP compressor) that is likewise not shown, and a low-pressure turbine portion (LP turbine) 16. This LP turbine comprises a plurality of stages, each stage 5 comprising a set 18 of stationary vane inserts 20 secured to an outer stator 22 together with a wheel 24 of moving blade inserts 40 connected to a rotor 26. The rotor 26 is coupled to a shaft (not shown) for rotating a low-pressure compressor (LP compressor), likewise not shown.

10 As can be seen more clearly in Figure 2, each stationary vane insert 20 comprises a vane 30, an outer platform 32 whereby the vane is connected to the stator 22, and an inner platform 34. The inner platform is provided with an abradable coating 36 facing sealing 15 wipers 38 carried by the rotor 26.

According to a characteristic of the invention, each moving blade insert 40 has a plurality of blades. In the example shown in Figure 3, each moving blade insert 40 comprises two blades 42, 44. The ends of the blades 42, 20 44 are interconnected by an outer platform 46 and an inner platform 48 they share in common. A root 60 common to the blades 42 and 44 projects from the inside of the platform 48 for connecting the insert 40 to the rotor 26.

On the outside, the outer platform 46 presents 25 sealing wipers 50 facing an abradable coating 52 carried by the stator 22 (Figure 2).

In the embodiment shown, the root 60 is a hollow root that is bulb-shaped. The root 60 presents a blind recess 62 formed in a portion of its thickness, as can be 30 seen in Figure 4, the recess 62 is open to one side 60a of the root, and specifically the side that is downstream (in the gas flow direction in the turbine). The recess also opens out via an orifice 63 in the base of the root 60. As a result, a flow of air can be established in the 35 recess 62, thereby contributing to cooling the root and the inner platform of the insert. The hollow root can be obtained directly by casting the blade insert. The

recess can alternatively be made by machining a solid root.

The connection between the blade insert 40 and the rotor 26 is achieved by engaging the root 60 in a housing 5 of complementary shape formed in the periphery of a rotor disk. Once the inserts have been assembled to the rotor so as to form a complete wheel, the outside faces of the inner platforms 48 make up a continuous inside surface for the gas flow into which the blades 42, 44 extend. 10 Compared with a wheel comprising the same number of blades, but in which the blade inserts have only one blade each, the number of gaps between adjacent inserts is halved. This enables the performance of the turbine to be improved by reducing leaks between insert 15 platforms. Sealing at the junctions between the inner platforms of pairs of adjacent inserts can be improved by means of a deflector-forming piece 54 (Figure 5). The piece 54 is an annular sector of upside-down channel section bridging the junction between two adjacent inner 20 platforms, and coming into contact with the inside faces of the end portions of said platforms, the flanges of the piece 54 pressing against the facing flanks of the roots 60 of two adjacent blade inserts.

In addition, the outer platform 46 of the insert 40 25 extends over a circumferential direction that is greater than that of the outer platform of an insert having a single blade only. This makes it possible to implement one or more sealing wipers 50 that extend continuously over a greater distance in the circumferential direction, 30 thereby making it possible to improve sealing between the tips of the moving blade insert 40 and the abradable coating facing them on the stator 22.

According to a particular characteristic of the invention, the resonant frequency of a moving blade 35 insert 40 is adjusted by acting on the configuration of the insert root.

With a hollow root 60 of the kind shown by way of example in Figures 3 and 4, the resonant frequency can be adjusted by adjusting the thickness of material at the neck 64, i.e. the most constricted portion of the root, 5 which thickness amounts to the sum of the thicknesses 64a and 64b (Figure 4) on either side of the recess. Thickness can be adjusted by acting in one direction or the other on the width of the recess 62 level with the neck 64, which has no influence on the outside dimensions 10 of the root 60. The thicknesses 64a and 64b are not necessarily equal. Quite a broad range of options is thus made available for adjusting the resonant frequency of the blade without interfering with the aerodynamic behavior or the size of the blade insert.

15 It should also be observed that making a hollow root contributes to reducing the weight of the moving blade insert, while also opening up the possibility of cooling by having an open cavity in the vicinity of the inner platform, with this being achieved without interfering 20 with mechanical performance. In the example shown, mechanical performance, and in particular ability to withstand the stresses exerted on the moving blade insert, is mainly a function of the thickness of the inner platform and of the flanks 66a and 66b of the root 25 60, and also of the base 68 of the root 60.

In Figure 6, curve I plots the resonant frequency F of a moving blade insert in the LP turbine of an aviation turbojet, of the kind shown in Figures 3 and 4, as a function of the speed of rotation of the LP turbine. 30 This frequency F can be adjusted by adjusting the thickness of material in the neck of the hollow root. In the same graph, curve $N_2 - N_1$ represents the main excitation mode at the lowest frequencies where N_2 and N_1 are the speeds of rotation of the HP and LP turbines, 35 respectively.

By way of comparison, curve II plots the resonant frequency F' of a prior art single-blade moving insert

having a blade of the same dimensions and shape as each of the two blades in the insert whose resonant frequency is equal to F , the single-blade insert and the two-blade insert being designed to build up interchangeable turbine 5 wheels having the same dimensions.

It can be seen that at a high speed of rotation, the resonant frequency F' presents a value that is only 14% greater than the value corresponding to the $N_2 - N_1$ excitation mode. Unfortunately, it can be seen that 10 increasing the resonant frequency F' so as to take it further away from the $N_2 - N_1$ excitation mode is not possible in practice without interfering with the mechanical and aerodynamic behavior of the single-blade insert.

15 In contrast, by using a two-blade insert, it is possible, by acting on the configuration of the root that is common to both blades, to adjust the resonant frequency so as to give it a value that is far enough away from the frequency of the $N_2 - N_1$ excitation mode to 20 avoid any risk of undesirable vibration, and in particular a value that is more than 14% greater than the $N_2 - N_1$ mode, and is preferably more than 20% greater. Thus, as shown in Figure 6, the resonant frequency F obtained with the two-blade insert used can be adjusted 25 to have a value that is more than 35% greater than the $N_2 - N_1$ excitation mode, i.e. more than 21% greater than the frequency of the single-blade insert.

In addition, according to another characteristic of the invention, the latitude available for adjusting the 30 resonant frequency of the moving inserts enables this frequency to be given values that are significantly different for the inserts belonging to the same moving wheel and/or for inserts belonging to different moving 35 wheels making up the LP turbine. Resonant frequencies are said herein to be significantly different when the values of those frequencies differ by at least 1%. Advantageously, the various frequencies are adjusted to

values that are more than 14% greater and preferably more than 20% greater than the low frequency $N_2 - N_1$ excitation mode, as defined above.

Thus, in at least some of the moving wheels, and 5 preferably in all of the moving wheels forming an LP turbine stage, it is possible to use inserts of different resonant frequencies so as to ensure that the moving wheel is out of tune. Various different resonant frequencies are achieved by giving different shapes or 10 dimensions to the recesses in the roots. It is also possible in a single wheel to assemble blade inserts having hollow roots in association with blade inserts having solid roots without any recesses. In order to limit the number of different parts, it is possible in 15 any one wheel to make use of only a few different configurations of insert roots. This serves to avoid vibration occurring simultaneously in all of the inserts of a wheel.

It is also possible to give the blade inserts of one 20 moving wheel a resonant frequency that is different from that of the blade inserts in another moving wheel so as to ensure that the different moving wheels are out of tune with each other. Resonant frequency can be varied from one wheel to another by acting in particular on the 25 shapes and/or dimensions of the recesses in the insert root. It is also possible to make some moving wheels with blade inserts having solid roots without any recesses. This serves to avoid vibration appearing simultaneously in all of the moving wheels of a turbine.

30 The two above possibilities can be combined, i.e. with moving wheels in which at least some of them are out of tune at a set of resonant frequencies for their blade inserts that differ between different wheels.

Other modifications can also be made to the 35 embodiment described above without going beyond the ambit of the invention.

Thus, in another embodiment shown diagrammatically in Figure 7, the insert 140 differs from that of Figures 3 and 4 solely in that the root 160 of the insert 140 is Christmas-tree-shaped. Frequency can be adjusted 5 as described above by conferring a particular configuration to a recess 162 which is preferably blind and which is formed in the root 160. The root 162 may open out via an orifice 163 in the base of the root 160.

In another embodiment, the moving blade insert may 10 have more than two blades united between common outer and inner platforms and associated with a common root.

Although the above description relates to an LP turbine for a turbomachine, the invention can also be applied to the moving wheels of the HP turbines of 15 turbomachines. The invention can also be applied not only to aviation turbines, but also to industrial turbines.